#### Advancements in Gas Turbine Repair Technology Resulting from Gas Turbine Design Evolution

#### Scott Hastie, P.Eng Liburdi Turbine Services







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## Agenda

- Gas Turbine Design Evolution
- Repair Process
- Technological Advancements to Address Repair Challenges



#### **Design Evolution**



Westinghouse W191 Blade

GE Frame 7FA Row 1 Bucket



# Gas Turbine Design

- Performance Improvements
- Increased Power Output
- Increased Efficiency
- Reduced Emissions
- Increased Firing Temperatures
- Increased Flexibility
- Extended Maintenance Intervals



# Gas Turbine Design

- Material Improvements
- Advanced Coatings
- Advanced Cooling Designs
- Advanced Geometry



Frame 7FA



## **Design Evolution**





## F – Class Design

- F Class Design derived from Aero Blade
  - Internal cooling circuit design, alloy selection
  - Advanced technologies applied from GE, Rolls-Royce aero experience







## F – Class Design

Manufacturer/Model	First Stage Turbine Blade / Bucket
Alstom-GT26	Single Crystal, Ni-Base Matrix Cooled VPS-NiCoCrAIY with APS-TBC
GE-Frame 7FA	Directionally Solidified, GTD111 Serpentine Cooling w/Turbulators LPPS-CoCrAIY/DVC-TBC, Plus Internal Coatings
Siemens V94.3A	Single Crystal PWA1480 VPS-CoNiCrAIYSi Plus Internal Coatings
Siemens-Westinghouse 701G	Directionally Solidified CM247 Serpentine, Film & Showerhead VPS-NiCoCrAIY/TBC
GE LM6000 / RR Trent	Single Crystal, Serpentine, Film, Showerhead Internal coating, EBPVD TBC







Not only applicable To Blades





Frame 7FA



#### Gas Turbine Materials



#### **Turbine Rotating**

N105, N108, N115, Waspalloy, U-500, U520, U700,U710, U720, INX750, IN738, Rene80, GTD111, Mar-M247, Mar-M002, PWA1483, CMSX4, ReneN5

#### **Turbine Stationary**

300SS, 400SS, C242, C1023 N-155, M509, HS-188,L605 X-40, X-45, FSX-414, ECY-768 IN738, R80, GTD222, GTD444

#### Compressor

**Casings** Grey Cast Iron Carbon Steel Aluminum

#### **Turbine Shells**

Ductile Cast Iron Stainless Steel Nickel Alloy

Compressor Wheels/Disks Ni-Cr-MO-V Forging

#### Turbine Wheels/Discs Ni-Cr-MO-V Steel Cr-Mo-V Forging 12Cr Stainless Discalloy A286 IN718





## **Design Evolution - Cooling**





## **Design Evolution - Cooling**



After Frasier US Patent 5295530



## **Blade Repair Process**

**Incoming Inspection** 

Strip Coatings

Fluorescent Penetrant Inspection

**Dimensional Repair** 

Machine/Finish Geometry

**Rejuvenation Heat Treatment** 

**Pre-Coating Inspection** 

Coating

**Final Inspection** 







## **Incoming Inspection**

- Triage
  - Is the component repairable?
  - What is the expected level of repair?
- Solid Blades vs. F-Class Internal Geometry
- Increased reliance on life analysis of blade











### Life Analysis

DS Alloy Damage Cracking along DS grain boundaries at tip Oxidation burning at tip Coating cracks in airfoil #1 Cooling hole crack Incomplete internal coating











### **Incoming Inspection**

• Internal Geometry





## **Incoming Inspection**



## Internal Cleaning

- Internal Cleaning using thermal and chemical process are necessary before stripping
  - Internal deposits and oxides limit effectiveness of stripping process





## Internal Cleaning

• Oxide removal allows complete stripping



Before cleaning

After cleaning



# **Coating Stripping**

- Masking Required for Internal Only Strip
- Internal Stripping



Pump





## **Dimensional Repair**

- Tip weld of un-shrouded blades
- Z-notch restoration of shrouded blades
- Seal fin restoration





#### Weldability

#### Weldability of Superalloys





### **Dimensional Repair**

- Welding of gamma prime (γ<sup>´</sup>) strengthened superalloys is not as easy when compared to welding of cobalt base alloys or stainless steel alloys.
- Fusion Zone cracking
- HAZ cracking (microfissuring)
- Post weld heat treat cracking (strain age cracking









#### Materials – Single Crystal





#### **Material Selection**



New Blade





#### **Repaired Blades**





## Geometry Repair - Vanes

Liburdi Powder Metallurgy (LPM<sup>™</sup>) is a modified wide gap brazing process.

Can be used to fill very large gaps of more than 0.5" size

Build-up damaged surfaces

Up to 0.15" thick per application, multiple layers per repair

Heat treatments tailored to match the substrate alloy.

Component distortion greatly reduced using LPM<sup>™</sup> compared to conventional weld methodology.

Much stronger than braze repairs.

Nickel based LPM<sup>™</sup> materials can be applied to nickel, cobalt and stainless steel alloys.



Crack, crazed & oxidized surface



Blend to remove crack, crazing & oxidation



LPM<sup>™</sup> applied to surfaces



Heat treated and blended deposits



### Geometry Repair - Vanes

Original braze joint requires repair/restoration

•Incoming damage – burned leading and trailing edges , thin airfoils, burned outer shroud surfaces, shroud braze joints deteriorated

•Previously NGVs were considered not repairable and were replaced with new.

Severely burned trailing edges



#### **Geometry Repair - Vanes**

 LPM high strength alloy as-applied to leading edge, mid airfoil, and trailing edges

ED)

 LPM material after high temperature vacuum heat treatment to "cast" material to the airfoils

# Machining / Finishing

- Tip Cap Machining
- EDM Hole Drilling



**Cooling Hole Drilling** 



#### Heat Treatment

#### Liburdi Turbine Services Heat Treatment Development Timeline



## Coating

Selection of Coating System





# Coating

External airfoil surfaces – MCrAIY with over aluminize or MCrAIY bond coat with DVC or EB-PVD TBC.

Recoating internal airfoil surfaces – diffusion aluminide or silicon aluminide

- Handler March Charles



## Coating





#### **Repair Evolution**



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#### Thanks for Listening.



## Any Questions?

